

In the Abstract:

Please replace the paragraph at page 17, lines 2 to 15, with a replacement paragraph amended as follows:

~~The invention relates to an arrangement for detecting a shaft break in a rotor of a first turbine (10), particularly a medium pressure turbine, of a gas turbine, particularly of an aircraft engine, whereby~~ A gas turbine machine includes a second turbine (11), particularly e.g. a low pressure turbine, [[is]] positioned downstream of the from a first turbine (10), [[with]] e.g. a medium pressure turbine. An arrangement for detecting a shaft break of a rotor shaft includes an operator element (16) positioned between the rotor of the first turbine (10) and a stator of the second turbine (11) radially inwardly relative to a flow channel, and [[with]] a sensor element (21) guided in the stator of the second turbine (11), in order to convert a shaft break, detected by the radially inwardly positioned operator element (16), into an electrical signal and to transmit this electrical signal to a switching element which is positioned radially outwardly relative to the flow channel on a housing of the gas turbine machine.